

EML 4230 Introduction to Composite Materials

Chapter 2 Macromechanical Analysis of a Lamina **2D Stiffness and Compliance Matrix for Unidirectional Lamina**

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Courtesy of the Textbook

[Mechanics of Composite Materials by Kaw](#)



Compliance and Stiffness Matrix Elements in Terms of Elastic Constants

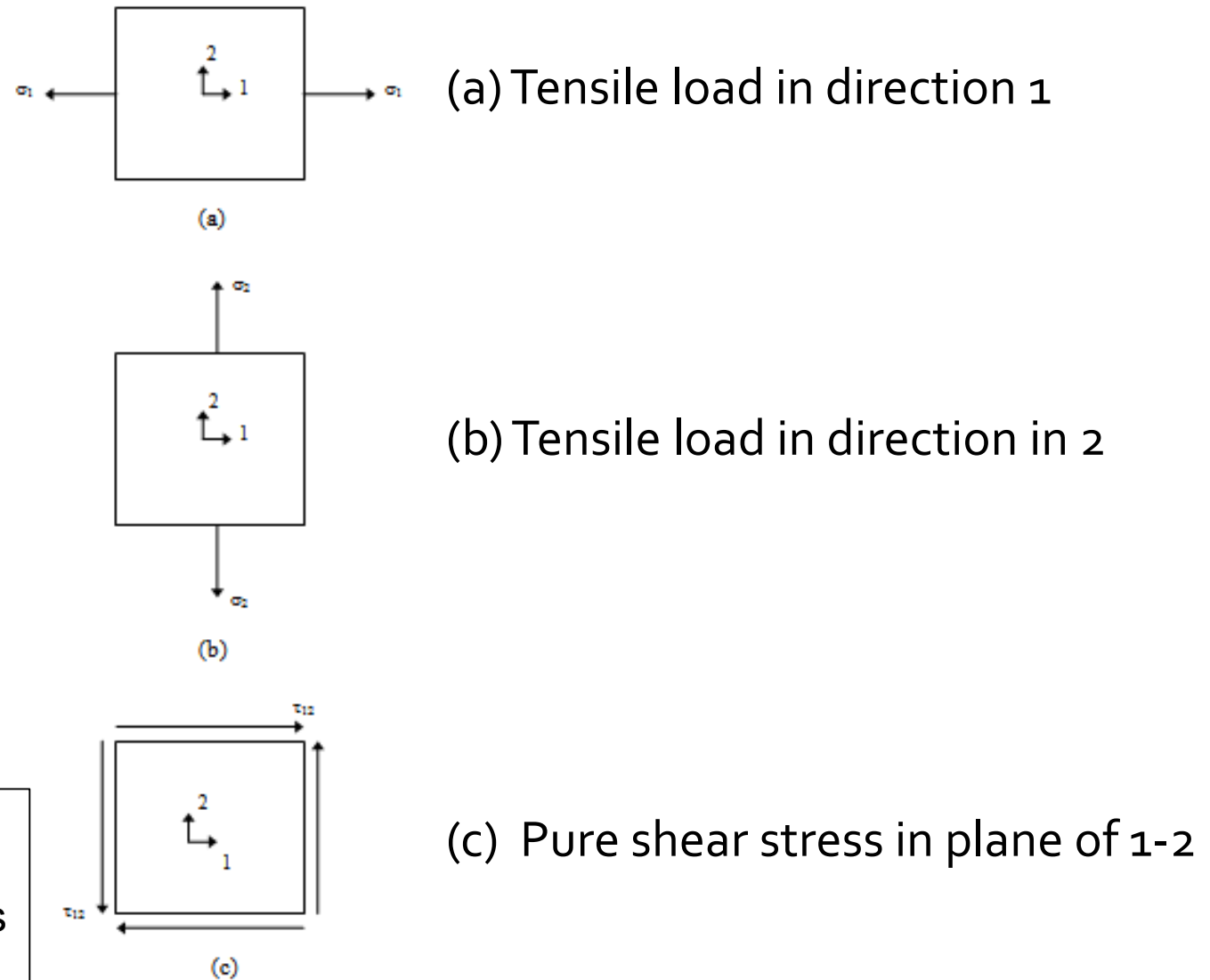
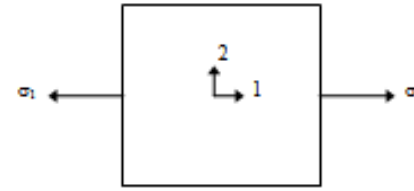


FIGURE 2.18
Application of stresses to
find engineering constants
of a unidirectional lamina

Pure Axial Load in Direction 1

Apply a pure axial load in direction 1

$$\sigma_1 \neq 0, \sigma_2 = 0, \tau_{12} = 0$$



$$\begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} S_{11} & S_{12} & 0 \\ S_{12} & S_{22} & 0 \\ 0 & 0 & S_{66} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} \quad \longrightarrow \quad \begin{aligned} \varepsilon_1 &= S_{11} \sigma_1 \\ \varepsilon_2 &= S_{12} \sigma_1 \\ \gamma_{12} &= 0 \end{aligned}$$

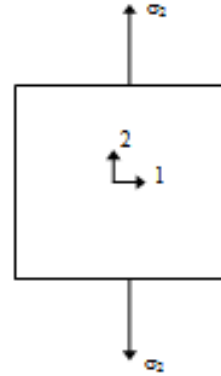
$$E_1 \equiv \frac{\sigma_1}{\varepsilon_1} = \frac{1}{S_{11}} \quad \longrightarrow \quad S_{11} = \frac{1}{E_1}$$

$$\nu_{12} \equiv -\frac{\varepsilon_2}{\varepsilon_1} = -\frac{S_{12}}{S_{11}} \quad \longrightarrow \quad S_{12} = -\frac{\nu_{12}}{E_1}$$

Pure Axial Load in Direction 2

Apply a pure axial load in direction 2

$$\sigma_1 = 0, \sigma_2 \neq 0, \tau_{12} = 0$$

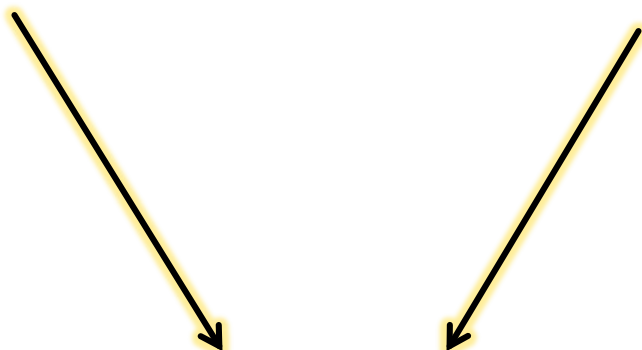


$$\begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} S_{11} & S_{12} & 0 \\ S_{12} & S_{22} & 0 \\ 0 & 0 & S_{66} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} \longrightarrow \begin{aligned} \varepsilon_1 &= S_{12} \sigma_2 \\ \varepsilon_2 &= S_{22} \sigma_2 \\ \gamma_{12} &= 0 \end{aligned}$$

$$E_2 \equiv \frac{\sigma_2}{\varepsilon_2} = \frac{1}{S_{22}} \longrightarrow S_{22} = \frac{1}{E_2}$$

$$\nu_{21} \equiv -\frac{\varepsilon_1}{\varepsilon_2} = -\frac{S_{12}}{S_{22}} \longrightarrow S_{12} = -\frac{\nu_{21}}{E_2}$$

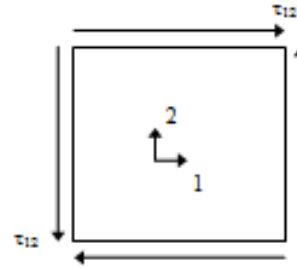
Reciprocal Relationship

$$S_{12} = -\frac{v_{12}}{E_1}$$
$$S_{12} = -\frac{v_{21}}{E_2}$$

$$\frac{v_{12}}{E_1} = \frac{v_{21}}{E_2}$$

Pure Shear Load in Plane 12

Apply a pure shear load in Plane 12

$$\sigma_1 = 0, \sigma_2 = 0, \tau_{12} \neq 0$$



$$\begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} S_{11} & S_{12} & 0 \\ S_{12} & S_{22} & 0 \\ 0 & 0 & S_{66} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} \longrightarrow \begin{aligned} \varepsilon_1 &= 0 \\ \varepsilon_2 &= 0 \\ \gamma_{12} &= S_{66} \tau_{12} \end{aligned}$$

$$G_{12} \equiv \frac{\tau_{12}}{\gamma_{12}} = \frac{1}{S_{66}} \longrightarrow S_{66} = \frac{1}{G_{12}}$$

Compliance Matrix

$$\begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} S_{11} & S_{12} & 0 \\ S_{12} & S_{22} & 0 \\ 0 & 0 & S_{66} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix}$$

$$\begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} \frac{1}{E_1} & -\frac{\nu_{12}}{E_1} & 0 \\ -\frac{\nu_{12}}{E_1} & \frac{1}{E_2} & 0 \\ 0 & 0 & \frac{1}{G_{12}} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix}$$

Coefficients of Stiffness Matrix

$$\begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} = \begin{bmatrix} Q_{11} & Q_{12} & 0 \\ Q_{12} & Q_{22} & 0 \\ 0 & 0 & Q_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix}$$



$$Q_{11} = \frac{S_{22}}{S_{11}S_{22} - S_{12}^2}$$

$$Q_{12} = -\frac{S_{12}}{S_{11}S_{22} - S_{12}^2}$$

$$Q_{22} = \frac{S_{11}}{S_{11}S_{22} - S_{12}^2}$$

$$Q_{66} = \frac{1}{S_{66}}$$



$$\begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} = \begin{bmatrix} \frac{E_1}{1 - \nu_{21}\nu_{12}} & \frac{\nu_{12} E_2}{1 - \nu_{21}\nu_{12}} & 0 \\ \frac{\nu_{12} E_2}{1 - \nu_{21}\nu_{12}} & \frac{E_2}{1 - \nu_{21}\nu_{12}} & 0 \\ 0 & 0 & G_{12} \end{bmatrix} \begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix}$$

END